

Control of collision orbits

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In the n -body problem, the forces acting between particles approach infinity when the mutual distances approach zero. Therefore, at collision, the equations of motion have singularities. Since Levi-Civita and Sundman, the double collision has been "regularized", i.e. the singularity has been made to disappear by means of algebraic transformations. Levi-Civita obtained first a regularizing transformation of the Kepler problem based on the map $z \rightarrow z^2$ of the complex plane. This conformal map sends the orbits of the harmonic oscillator in the ones of the Kepler problem. The coordinates transformation is coupled with a slowing down of the motion by means of a time change defined by $d\tau = 1/|z|dt$.

Based on classical results, the purpose of the work is the application of the regularization theory to optimal control. We consider the control of a spacecraft under the attraction of one or more bodies, where the control is the thrust, and with the goal of minimizing the integral of some function f . If $f = 1$, the setting is a time-minimization problem. The *Pontryagin's maximum principle (PMP)* allows us to reduce the differential system to the analysis of an Hamiltonian system for finding optimal solutions. However, the system exhibits singularities in zero, and thus the *PMP* fails in the study of collision orbits. By applying the Levi-Civita regularization to the controlled system, we obtain an affine system (in the new time τ) where the vector fields are complete and C^∞ .

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